

**IN THE SPECIFICATION:**

Please replace the paragraph at page 7, lines 17-22, with the following paragraph:

The blade 10 of Figure 1 generally includes an airfoil 12 against which hot combustion gases are directed during operation of the gas turbine engine, and whose surface is therefore subject to severe attack by oxidation, corrosion and erosion. The airfoil 12 is anchored to a turbine disk (not show) with a dovetail 14 formed on a platform 16 of the blade 10. Cooling holes [18] 18' are present in the airfoil 12 through which bleed air is forced to transfer heat from the blade 10.

Please replace the paragraph at page 16, lines 12-19, with the following paragraph:

The difference in thickness,  $\Delta x$ , between the prior bond coat 20 thickness and new bond coat 21 also may be determined. This difference,  $\Delta x$ , may be determined by, for example, knowing the actual value for  $x_{NiAl}$  and known value of  $x_{add}$  and determining the difference between them (e.g.,  $\Delta x = x_{NiAl} - x_{add}$ ). If unknown,  $x_{add}$  may be approximated at about 1.2 mil. The ceramic thermal barrier coating 22 may then be applied at the newly determined nominal thickness of  $t + \Delta t - \Delta x$ . If, for instance,  $x_{NiAl} > x_{add}$ , the preferred TBC thickness will be less than  $t + \Delta t$ , and if  $x_{NiAl} [ \leq ] \leq x_{add}$ , the final intended TBC thickness will be more than  $t + \Delta t$ .